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ИЗВЕСТИЯ

РОО «НАЦИОНАЛЬНОЙ
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NAS RK is pleased to announce that News of NAS RK. Series of geology and technical sciences scientific journal has been accepted for indexing in the Emerging Sources Citation Index, a new edition of Web of Science. Content in this index is under consideration by Clarivate Analytics to be accepted in the Science Citation Index Expanded, the Social Sciences Citation Index, and the Arts & Humanities Citation Index. The quality and depth of content Web of Science offers to researchers, authors, publishers, and institutions sets it apart from other research databases. The inclusion of News of NAS RK. Series of geology and technical sciences in the Emerging Sources Citation Index demonstrates our dedication to providing the most relevant and influential content of geology and engineering sciences to our community.

Қазақстан Республикасы Ұлттық ғылым академиясы «ҚР ҰҒА Хабарлары. Геология және техникалық ғылымдар сериясы» ғылыми журналының Web of Science-тің жаңаланған нұсқасы Emerging Sources Citation Index-те индекстелуге қабылданғанын хабарлайды. Бұл индекстелу барысында Clarivate Analytics компаниясы журналды одан әрі the Science Citation Index Expanded, the Social Sciences Citation Index және the Arts & Humanities Citation Index-ке қабылдау мәселесін қарастыруда. Web of Science зерттеушілер, авторлар, баспашылар мен мекемелерге контент тереңдігі мен сапасын ұсынады. ҚР ҰҒА Хабарлары. Геология және техникалық ғылымдар сериясы Emerging Sources Citation Index-ке енуі біздің қоғамдастық үшін ең өзекті және беделді геология және техникалық ғылымдар бойынша контентке адалдығымызды білдіреді.

НАН РК сообщает, что научный журнал «Известия НАН РК. Серия геологии и технических наук» был принят для индексирования в Emerging Sources Citation Index, обновленной версии Web of Science. Содержание в этом индексировании находится в стадии рассмотрения компанией Clarivate Analytics для дальнейшего принятия журнала в the Science Citation Index Expanded, the Social Sciences Citation Index и the Arts & Humanities Citation Index. Web of Science предлагает качество и глубину контента для исследователей, авторов, издателей и учреждений. Включение Известия НАН РК. Серия геологии и технических наук в Emerging Sources Citation Index демонстрирует нашу приверженность к наиболее актуальному и влиятельному контенту по геологии и техническим наукам для нашего сообщества.

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ROUTE CONTROL AND COLLISION AVOIDANCE FOR MULTIPLE UAVS IN A SMART CITY CONTEXT USING GEOGRAPHIC INFORMATION SYSTEM

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Abstract: Unmanned Aerial Vehicles (UAVs) are complex hardware and software systems with multi-purpose control, playing a crucial role in implementing the smart city concept. Unlike manned aviation, UAV control systems require additional modules, including the drone itself, operator workstation, software, data transmission channels, and units necessary for fulfilling assigned tasks in urban

environments. The application of UAVs in the civilian sector, especially in the context of smart cities, is virtually unlimited. However, the current state of the legal framework for airspace usage creates certain difficulties for flight operations. The article formulates the main scientific position on the methodology for solving auxiliary tasks, defines research stages, and proposes a generalized methodological algorithm that includes theoretical developments, field observations, and computer modeling of smart city scenarios. The study revealed that UAV motion control systems are used for their differentiation process based on the principle of complete external control. For external control of UAV divergence processes in urban spaces, a method is considered for assessing UAV convergence situations and selecting divergence maneuvers using the area of dangerous courses. This takes into account the inertia of UAVs during turns and the presence of navigational hazards in the smart city maneuvering zone.

Keywords: UAVs, Geographic Information System (GIS), smart city, route management, collision avoidance, urban airspace, motion control system, maneuvering, navigational hazards, flight simulation, control automation, spatial analysis, air traffic monitoring, route optimization, flight safety.

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ГЕОАҚПАРАТТЫҚ ЖҮЙЕНІ ҚОЛДАНА ОТЫРЫП, АҚЫЛДЫ ҚАЛА КОНТЕКСТІНДЕ МУЛЬТИ ҰШҚЫШСЫЗ ҰШУ АППАРАТТАРЫ ҮШІН БАҒЫТТАРДЫ БАСҚАРУ ЖӘНЕ СОҚТЫҒЫСУДЫ БОЛДЫРМАУ

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Аннотация: Ұшқышсыз ұшу аппараттары (ҰҰА) - бұл көп мақсатты басқару жүйесі бар күрделі аппараттық-бағдарламалық кешендер, олар ақылды қала тұжырымдамасын жүзеге асыруда маңызды рөл атқарады. Пилотты авиациядан айырмашылығы, ҰҰА басқару жүйелері қосымша модульдерді қажет етеді, оның ішінде дронның өзі, оператордың жұмыс орны, бағдарламалық жасақтама, деректерді беру арналары және қалалық ортада белгіленген міндеттерді орындау үшін қажетті блоктар бар. ҰҰА -ны азаматтық секторда, әсіресе ақылды қала контекстінде қолдану іс жүзінде шектеусіз. Алайда, әуе кеңістігін пайдаланудың қазіргі құқықтық базасының жағдайы ұшу операцияларын жүргізуде белгілі бір қиындықтар тудырады. Мақалада қосалқы міндеттерді шешу әдістемесі бойынша негізгі ғылыми ұстаным тұжырымдалады, зерттеу кезеңдері анықталады және теориялық әзірлемелерді, далалық бақылауларды және ақылды қала сценарийлерін компьютерлік модельдеуді қамтитын жалпыланған әдіснамалық алгоритм ұсынылады. Зерттеу ҰҰА қозғалысын басқару жүйелері толық сыртқы басқару қағидаты бойынша оларды саралау процесі үшін қолданылатынын көрсетті. Қалалық кеңістіктегі ҰҰА ыдырау процесін сыртқы бақылау үшін қауіпті бағыттар аймағын қолдана отырып, ҰҰА жақындасу жағдайын бағалау және олардың ыдырау маневрін таңдау әдісі қарастырылады. Бұл ретте ақылды қаланың маневр аймағындағы ПҰА-ның бұрылу кезіндегі инерциясы мен навигациялық қауіптердің болуы ескеріледі.

Түйін сөздер: ҰҰА, геоаппараттық жүйе (ГАЖ), ақылды қала, бағыттарды басқару, соқтығысуды болдырмау, қаланың әуе кеңістігі, қозғалысты басқару жүйесі, маневр жасау, навигациялық қауіптер, ұшуды модельдеу, басқаруды автоматтандыру, кеңістіктік талдау, әуе қозғалысын бақылау, бағыттарды оңтайландыру, ұшу қауіпсіздігі.

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УПРАВЛЕНИЕ МАРШРУТАМИ И ПРЕДОТВРАЩЕНИЕ СТОЛКНОВЕНИЙ ДЛЯ МУЛЬТИ-БПЛА В КОНТЕКСТЕ УМНОГО ГОРОДА С ПРИМЕНЕНИЕМ ГЕОИНФОРМАЦИОННОЙ СИСТЕМЫ

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Аннотация. Беспилотные летательные аппараты (БПЛА) представляют собой сложные аппаратно-программные комплексы с многоцелевым управлением, играющие важную роль в реализации концепции умного города. В отличие от пилотируемой авиации, системы управления БПЛА требуют дополнительных модулей, включающих сам дрон, рабочее место оператора, программное обеспечение, каналы передачи данных и блоки, необходимые для выполнения поставленных задач в городской среде. Применение БПЛА в гражданском секторе, особенно в контексте умного города, практически не ограничено, однако текущее состояние правовой базы использования воздушного пространства создает определенные трудности для полетных операций. В статье формулируется основная научная позиция по методологии решения вспомогательных задач, определяются

этапы исследования и предлагается обобщенный методологический алгоритм, включающий теоретические разработки, полевые наблюдения и компьютерное моделирование сценариев для умного города. Исследование показало, что системы управления движением БПЛА используются для процесса их дифференциации по принципу полного внешнего управления. Для внешнего контроля процесса расхождения БПЛА в городском пространстве рассматривается метод оценки ситуации сближения БПЛА и выбора маневра их расхождения с использованием области опасных курсов. При этом учитывается инерция БПЛА при повороте и наличие навигационных опасностей в зоне маневрирования умного города.

Ключевые слова: БПЛА, геоинформационная система (ГИС), умный город, управление маршрутами, предотвращение столкновений, воздушное пространство города, система управления движением, маневрирование, навигационные опасности, моделирование полетов, автоматизация управления, пространственный анализ, мониторинг воздушного движения, оптимизация маршрутов, безопасность полетов.

Introduction. During the control of the aircraft movement, aerodynamic forces and moments arise (Woo G.S, 2020). The angles of deviation in pitch, yaw, roll and thrust of the engine are used as regulatory factors that allow influencing the aircraft to control its movement. An unmanned aerial vehicle (UAV) as a control object is a complex dynamic system due to the presence of a large number of interconnected parameters and complex cross-interactions between them (Sahawneh L.R., 2018). Complex motion is often divided into the simplest types: angular motion and the movement of the centre of mass, longitudinal and lateral motion (Sahawneh L.R., 2015). The controls that create control actions can be divided into two groups: longitudinal controls that provide movement in the longitudinal plane; lateral motion controls that provide the required nature of changes in the angles of roll, slip and yaw (Rattanagraikanakorn B., 2019). The engine control channel regulates thrust in accordance with the specified flight program (Highland P., 2020; Batrakov D.O, 2016). Stable flight control is impossible without creating an acceptable quality automatic control system (Minwalla C., 2016; Kang C., 2020). For example, the aircraft control system serves to ensure flight along a given trajectory by creating the necessary aerodynamic forces and moments on the wing and fins (Clothier R. A., 2015). Three types of control systems are possible – manual, semi-automatic and automatic (Orazbayev B, 2020). In the manual control system, the pilot operator, assessing the situation, ensures the generation of control impulses and, using command levers through the control panels, deflects the steering surfaces, holding them in the desired position (Orazbayev B, 2023). In a semi-automatic system, the pilot operator control signals are transformed and amplified by various automata and amplifiers, providing optimal stability and controllability characteristics of the aircraft (Lee K., 2021; Tanirbergenova A, 2021). Automatic systems provide full automation of individual flight stages, freeing the pilot operator from direct

involvement in the control of the aircraft (Cherunova I., 2018). Swarm monitoring of UAVs is an integral part of a modern geographic information system, providing comprehensive collection, analysis, and visualization of spatial data for effective urban environment management and collision avoidance in the airspace of a smart city.

In the process of adjusting the control by the angles or height of the aircraft flight in the automatic system, the desired values of angles or height are received at the input of the controller, and the output variables of the controller will deflect the angles of the ailerons along the pitch, roll and yaw channels (Cherunova I., 2018; Molloy T., 2020). The task of the synthesis of the aerobatic system is to select the structure and parameters of the control channels that provide a given quality of flight control, based on dynamic properties. Actuators are an integral part of automatic UAV motion control systems (BatrakovD., 2017; D'AmatoE., 2020). The inclusion of mathematical models of these devices in the control object allows taking into account their dynamic and static properties (Orazbayev B, Zhumadillayeva A., 2023). Steering actuators are selected from the condition that their load characteristics provide the necessary dynamics of control processes. In other words, they are required to ensure that the steering body loaded with external forces or external moments moves at a given speed (Orazbayev B., Zhumadillayeva A., 2023). The purpose of the study is to ensure the implementation of the requirements for the control system: the choice of a transition process with minimal time, the absence of overshoot (aperiodic process). It is necessary that the control system provides the specified parameters of the transition process. The goal set in this paper requires the study of the following tasks:

- justification of the mathematical description of the control object (CO);
- building a simulation model;
- study of the dynamics of the model;
- implementation of the laws of management of the management object.

Materials and methods. It is necessary to find analytical expressions for the boundaries of the area of dangerous courses and UAV speeds, taking into account the ratio of their speeds, based on the condition of equality of the distance of UAV's closest approach and the maximum permissible approach distance. Since the distance of the closest approach of the UAV depends on the course of the first UAV and on the speed of the second UAV, which are the parameters of the divergence manoeuvre, and the maximum permissible distance of approach is a constant, the relationship between the course of the first UAV and the speed of the second UAV is derived from the obtained equality, which is the equation of the limits of the area of dangerous courses and UAV speeds.

The area obtained in this way is used to determine a safe joint divergence manoeuvre if the turn and braking time of the UAV is insignificant and can be neglected. In the case of a significant braking time of the second UAV, a procedure for the area development should be developed, and the development algorithm should differ from the previous one. To take into account the influence of the inertial

braking characteristics of the second UAV on the possibility of divergence in the selected maximum permissible approach distance, it is necessary to use analytical expressions of the characteristics of the processes of active and passive braking.

The next stage of the work is the development of a method for accounting for the third UAV, which may interfere with the safe distinction of the first and second UAV. To solve the next stage, an analytical procedure should be developed to determine the distances of the closest approach of the third UAV with the first and second UAVs during their divergence. If these distances are greater than the maximum permissible approach distance, then a compatible manoeuvre selected using the area of dangerous values of the course of one UAV and the speed of the other is acceptable, since it ensures the safe divergence of all three UAVs. Then it is necessary to develop a way to visualize the developed area of dangerous values of the course and speed of the UAV and to form a verbal algorithm for determining the parameters of a safe manoeuvre of divergence at the selected point of the boundary of the developed area.

Results and discussions. To ensure the safety of the difference, the coordination of the UAV divergence manoeuvres is necessary, that is, the coordination of manoeuvres, which allows increasing the distance of the closest approach. With this type of control, each of the UAVs controls the current state of the approach situation and when a situational disturbance occurs, Bz interaction occurs between the UAVs, which transforms the programme section of relative motion with situational disturbance $D_{miny} \geq D_d$ into a relative trajectory without situational disturbance $D_{miny} \geq D_d$. The Bz interaction predicts the behaviour of UAVs in case of divergence and ensures the development of coordinated targeted strategies for each of the interacting UAVs. Therefore, the interaction of Bz can formally be written in this way:

$$G = Bz(F) \quad (1)$$

where $F = (D, \alpha, V_1, K_1, V_2, K_2)$ – is the vector of the convergence situation, $G = (G_1, G_2)$.

Thus, the interaction of the UAV Bz is the operator or the display of the parameters of the state of the approach situation in the set of parameters of the strategy difference G, and the interaction of consists of Bz two operators: Crd – coordination of manoeuvres and Prm – calculation of manoeuvre parameters.

The interaction of Bz , as a mechanism of coordination to achieve the common goal of preventing dangerous convergence, indicates the behaviour of each of the UAVs in the process of divergence, and a change in the situation is predicted, which is an extremely important factor affecting the safety of the difference. Thus, the process of differentiation is the process of compensating situational disturbance, that is, transferring the situation of convergence into a subset of safe states, according to the interaction mechanism Bz , and the strategy of difference G is the algorithm for

implementing the process of divergence. The implementation of the **Bz** interaction is carried out using a binary coordination system or a $c_0(Bz)$ coordinator, the input of which is a vector **F**, and the output is the address signals of the UAV β_1 and β_2 (Figure 1).

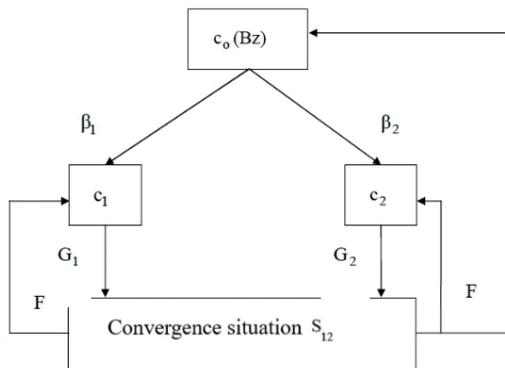


Figure 1. Independent control of the S_{12} system

With the help of the address signal β_i and the state vector **F**, each of the UAVS c_i of the S_{12} system selects a manoeuvre of divergence G_i from the permissible subset of evasion courses, which is regulated by the coordinating signal β_i . The $c_0(Bz)$ binary coordination system is a means for describing the behaviour trend of a pair of interacting UAVs when a situational disturbance occurs in order to compensate for it. As a divergence manoeuvre, the manoeuvre of changing the course of the UAV is considered. First of all, the binary coordination system $c_0(Bz)$ must satisfy Ashby's law of necessary diversity, according to which the variety of available difference strategies must correspond to the variety of possible situational perturbations. Otherwise, the $c_0(Bz)$ system will not be able to compensate for situational disturbances, creating prerequisites for UAV collisions. This means that the system $c_0(Bz)$ must have at its disposal the potential possibility of compensating situational disturbance in all cases at ω_1 .

In conditions of reduced visibility, UAV coordination is not provided for in the MPPSS- 72 standard. The second type of control of the UAV divergence process is their complete control by an external UAV operator, who observes the state of the convergence process and, in the event of a situational disturbance, develops a common divergence strategy for both UAVs, translating a dangerous convergence into an unfulfilled state. This type of control can be both a workstation control system (WCS) and, crucially, an on-board information system with the same capabilities installed on each of the UAVs, which solves the problem of collective compensation of situational disturbance and implements the individual strategy obtained as a result of the solution. The UAV operator Ξ observes the state vector **F** of the situation **S** of 12 approaching UAVs and analyses the presence of a

situational disturbance, at the appearance of which the operator chooses the optimal strategy (G_1, G_2) . Manoeuvres G_1 and G_2 , as shown in Figure 2, are addressed to the UAVs c_1 and c_2 that implement them.

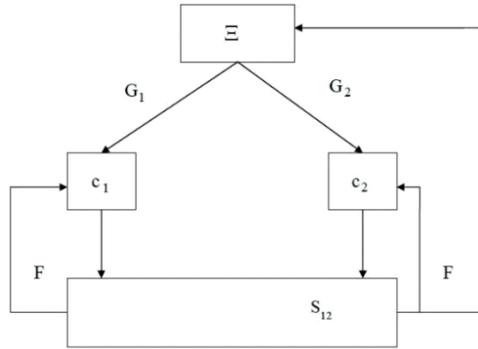


Figure 2. The principle of complete control of the S_{12} system.

The advantage of full control of the divergence process by the UAV operator by an external manager is the same interpretation of situational disturbance when choosing divergence manoeuvres G_1 and G_2 . Compensation of situational disturbance ω is developed in the first phase of evasion strategy G . After the completion of the evasion section, the exit section to the program trajectory of movement is realised. Recently, methods of external control of the UAV divergence process have been suggested, in particular, the paper considers the method of developing the area of dangerous UAV courses and procedures for assessing the situation of convergence and choosing evasion courses with its help. Here is a brief outline of the essence of the method of developing a dangerous area of UAV courses and its application. Let's assume that at the initial moment of time, the mutual position of two UAVs that are dangerously approaching is characterised by the bearing α_{ij} and the distance D_{ij} , and the mutual displacement is the relative course K_{otij} and the speed V_{otij} . The distance of the closest approach $\min D_{ij}$ of the UAV is less than the maximum permissible distance d_d , that is, $\min D_{ij} < d_d$, and the UAVs are approaching dangerously. The WCS, which controls the flight of the UAV, needs to find the courses of the UAV K_i and K_j , at which their distances of the closest approach will be greater than the maximum permissible distance d_d . For this purpose, we use the expression (1) for $\min D_{ij}$ and get:

$$\min D_{ij} = \Delta_{ij} D_{ij} \sin(\alpha_{ij} - K_{otij}) \geq d_d \tag{2}$$

The minimum approach distance $\min D_{ij}$ can be increased by changing the relative course K_{otij} , that is, the courses of the UAV K_i and K_j . Let's find the value of the UAV courses that ensure the fulfillment of the obtained condition, which can be represented as:

$$tgK_{otij} \leq tg \left[\alpha_{ij} - \text{arc sin} \frac{d_d}{\Delta_{ij} D_{ij}} \right] \quad (3)$$

Enter the designation:

$$\gamma_{ij} = \alpha_{ij} - \text{arc sin} \left(\frac{d_d}{\Delta_{ij} D_{ij}} \right) \quad (4)$$

and we take into account that according to:

$$tg K_{otij} = \frac{V_i \sin K_i - V_j \sin K_j}{V_i \cos K_i - V_j \cos K_j} \quad (5)$$

with this in mind, expression (23) takes the following form:

$$(\sin K_1 \cos \gamma_{ij} - \cos K_i \sin \gamma_{ij}) \leq p_{ij} (\sin K_j \cos \gamma_{ij} - \cos K_j \sin \gamma_{ij}) \quad (6)$$

where:

$$\rho_{III0} = V_j / V_i \quad (7)$$

Corresponding equality:

$$(\sin K_1 \cos \gamma_{ij} - \cos K_i \sin \gamma_{ij}) = p_{ij} (\sin K_j \cos \gamma_{ij} - \cos K_j \sin \gamma_{ij}) \quad (8)$$

or:

$$\sin(K_i - \gamma_{ij}) = \rho_{ij} [\sin(K_j - \gamma_{ij})] \quad (9)$$

represents analytical expressions of the boundary of the dangerous area S_{Dij} , which limits the invalid value combinations of the pairs of corresponding courses K_i and K_j .

The resulting equation has two solutions, that is, the following analytical dependencies are valid:

$$K_i - \gamma_{ij} = \text{arc sin} \{ \rho_{ij} [\sin(K_j - \gamma_{ij})] \} \quad (10)$$

$$K_i - \gamma_{ij} = \pi - \text{arc sin} \{ \rho_{ij} [\sin(K_j - \gamma_{ij})] \} \quad (11)$$

The first of the two solutions of the equation correspond to the situation of approaching UAVs, and the second corresponds to the situation of their removal. The danger of collision occurs when the UAV approaches, therefore, the equation of the boundaries of the dangerous areas of the courses S_{Dij} is determined by the equation:

$$K_i = \gamma_{ij} + \text{arc sin} \{ \rho_{ij} [\sin(K_j - \gamma_{ij})] \} \quad (12)$$

This expression characterizes the boundary between the dangerous and permissible areas of the courses K_i and K_j . Using the program, Figure 3 shows a graphical representation of the dangerous area S_{Dij} of the courses K_i and, in which $\min D_{ij} < d_d$, for a situation of dangerous approach with parameters $\alpha = 75^\circ$, $D = 3$ miles, $d_d = 1$ mile, $V_i = 15$ nodes, $V_j = 20$ nodes.

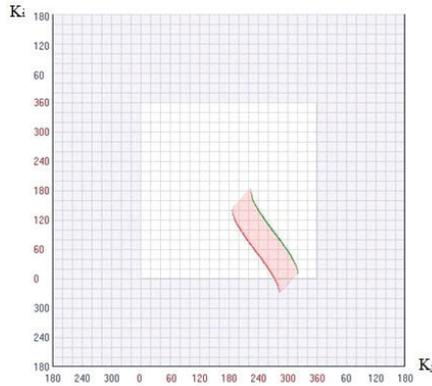


Figure 3. Dangerous area S_{Dij} of courses K_i and K_j at $V_i < V_j$

If the UAV speeds are equal, that is $V_i = V_j$, it follows:

$$K_i - \gamma_{ij} = \arcsin\{\sin(K_j - \gamma_{ij})\} \tag{13}$$

$$K_i - \gamma_{ij} = \pi - \arcsin\{\sin(K_j - \gamma_{ij})\} \tag{14}$$

The convergence condition corresponds to the dependence, the graphical form of which is shown in Figure 4.

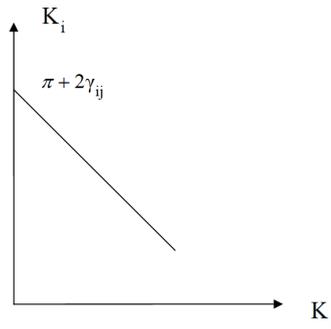


Figure 4. Dependence of the K_i course from K_j

Figure 5 shows the danger area S_{Dij} of courses K_i and K_j for the same situation, but with $V_i = 15$ nodes, $V_j = 15$ knots. Please note that when the UAV speeds are equal, the boundaries of the area of dangerous approaching courses S_{Dij} are straight lines. Figure 5 shows the danger area of courses and for the same situation, but with $V_i = 15$ nodes, $V_j = 15$ knots. Please note that when the UAV speeds are equal, the boundaries of the area of dangerous approaching courses are straight lines.

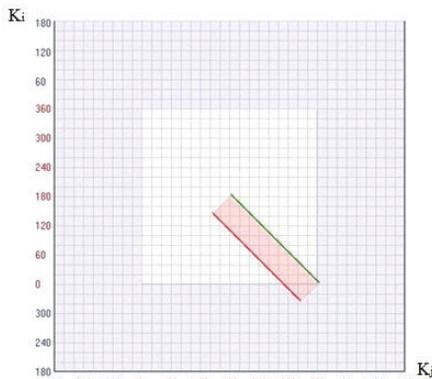


Figure 5. Dangerous area S_{Dij} of courses K_i and K_j at $V_i = V_j$

If $V_i > V_j$, that is, $\rho_{ij} < 1$, then the dependence between the courses of the UAV K_i and K_j for the case of convergence is expressed as follows:

$$K_i = \gamma_{ij} + \arcsin\{\rho_{ij}[\sin(K_j - \gamma_{ij})]\} \quad (15)$$

Since $\rho_{ij} < 1$, the course K_j takes all values from 0 to 2π . Figure 6 shows the danger area S_{Dij} of courses K_i and K_j for $V_i=20$ nodes, $V_j=15$ knots

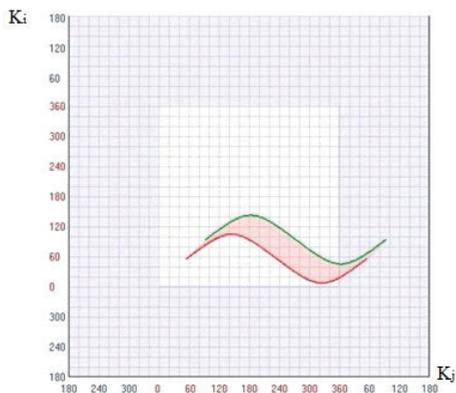


Figure 6. Dangerous area S_{Dij} of courses K_i and K_j at $V_i > V_j$

The assessment of the danger of the approach situation is carried out by analysing the position of the point of the UAV's initial courses (K_{ni}, K_{nj}) in relation to the dangerous area S_{Dij} . If $(K_{ni}, K_{nj}) \in S_{Dij}$, then $\min D_{ij} < d_a$ the situation of convergence is dangerous, otherwise there is no situational disturbance and there is no need to change the parameters of the movement of UAVs approaching. As an example, Figure 7 shows the danger area S_{Dij} for the following parameters: α_{ij}

$= 90^\circ$, $D_{ij} = 3.0$ miles, $d_d = 1.0$ mile, $V_i = 15$ nodes, $V_{ij} = 20$ nodes. For the initial courses $K_{ni} = 45^\circ$, and $K_{nj} = 315^\circ$, the starting point is located in the dangerous area of the courses (in Figure 7, the course K_j is counted along the ordinate axis and displayed by a horizontal line, and the course K_i is counted along the abscissa axis and displayed by a vertical line). For a given point, the distance of the closest approach is $\min D_{ij} = 0.42$ miles, so it is necessary to change the position of the point (K_i, K_j) by transferring its position from the dangerous area D_{ij} . For this purpose, a point M is selected that belongs to the boundary of the region S_{Dij} at the minimum distance from the starting point. The coordinates of this point are the values of the optimal UAV evasion courses. Figure 7 shows that the optimal values of the courses are $K_i = 63^\circ$, $K_j = 323^\circ$ – they are the coordinates of the point M, while $\Delta K_i = 18^\circ$ and $\Delta K_j = 8^\circ$. At the specified UAV courses, the distance of the closest approach $\min D_{ij} = 1.01$ miles, that is, equal to the maximum permissible distance.

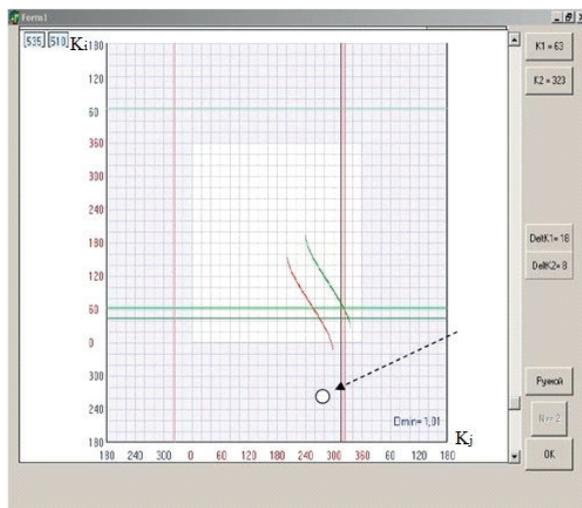


Figure 7. Determination of optimal values of courses K_i and K_j

Taking into account the inertia of the UAV when turning and navigational obstacles in the development of the area of dangerous courses and the choice of optimal evasion courses are considered in the work, and the inertia of the UAV when turning is taken into account by a dynamic rotation model with a constant angular velocity at a given rudder angle. Navigational hazards are taken into account by a set of unacceptable evasion courses, which is set by a pair of boundary unacceptable courses for each UAV. Thus, having at your disposal the dangerous area S_{Dij} of the courses of two UAVs, you can choose safe evasion courses that provide a discrepancy at a distance that is greater than the maximum permissible distance, while it is possible to take into account the inertia of the UAV and navigational obstacles.

In our work, it is shown that at the value of the difference between the courses of the UAV $\Delta K=0$ and $\Delta K=180$, that is, when the UAV approaches on “countercourses” and parallel courses, the change in the speeds of the UAV does not affect the value of the relative course. Therefore, in such situations, the UAV divergence manoeuvre by changing speeds is impossible, that is, many safe divergence manoeuvres are empty. The paper also indicates that when the current UAV speeds are equal, the ($\Delta V=0$) value of the relative course also does not change. Therefore, in the case of equality of the initial speeds with their identical change, the manoeuvre of divergence of the UAV by changing the speeds is also impossible.

It was shown above that the area of unacceptable speeds of a pair of UAVs has limits:

$$V_1^* = k * V_2, V_{1*} = k_* V_2 \tag{16}$$

where:

$$k^* = \frac{\sin(K_2 - \gamma^*)}{\sin(K_1 - \gamma^*)} \text{ and } k_* = \frac{\sin(K_2 - \gamma_*)}{\sin(K_1 - \gamma_*)} \tag{17}$$

Obviously, the existence of many safe manoeuvres of difference takes place under the condition:

$$\infty > k^* > 0, \infty > k_* > 0 \text{ and } k^* > k_* \tag{18}$$

Otherwise, the plural of safe divergence maneuvers is empty. The existing area of dangerous speeds has the form shown in Figure 8. In this example, the parameters of the convergence situation have the following values:

$$\alpha_0 = 45^\circ, D_0 = 3 \text{ miles}, D_d = 1 \text{ mile}, K_1 = 90^\circ, K_2 = 180^\circ \tag{19}$$

At initial speeds $V_1=18$ knots and $V_2=21$ knots, the distance of the shortest approach is $D_{min}=0.23$ miles.

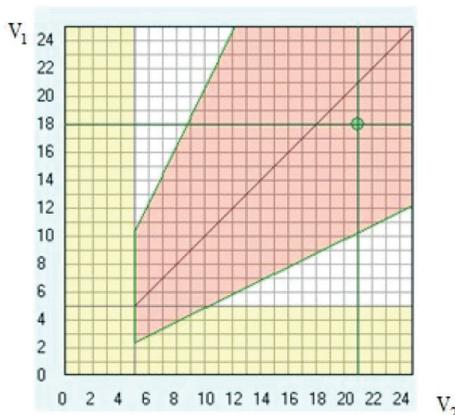


Figure 8. The area of dangerous UAV speeds

The above conditions are also valid for taking into account the inertial characteristics of the UAV when choosing a safe manoeuvre. In this case, to select their divergence rates, it has to be done as follows. A pair of safe speeds V_{1y} and V_{2y} of the area of dangerous speeds is determined. Due to the inertia of the UAV, the necessary speed differences V_{1y} and V_{2y} are achieved over time intervals in accordance with τ_{1y} and τ_{2y} , which are not equal to each other. Through this the total duration of the transition process t_p from the beginning of the UAV velocity change in the selected stable values V_{1y} and V_{2y} is equal to the larger of the intervals

$$\tau_{1y} \text{ and } \tau_{2y} (t_p = \max[\tau_{1y}, \tau_{2y}]) \quad (20)$$

This work is devoted to the second type of optimal divergence manoeuvre by reducing the speed, the feature of which is a fixed value of the moment of the beginning of the braking of the UAV, equal to the zero moments of time $t_n = 0$. The divergence manoeuvre is optimized according to the parameters V_{1y} and V_{2y} , which differ minimally from the initial values of the corresponding velocities V_1 , V_2 and ensure equality $D_{\min}(V_{1y}, V_{2y}) = D_d$ until the end of the transition process coinciding with the time of the closest approach.

To select the optimal manoeuvre of the second type, a programme has been developed that calculates the distance of the closest approach based on the entered values of the evasion speeds V_{1y} and V_{2y} , taking into account the selected braking modes, and allows you to determine the optimal values of the UAV evasion speeds.

As an example, the situation of a dangerous approach of a UAV with speeds of $V_1 = 17$ knots and $V_2 = 22$ knots is considered. As shown in Figure 9, the convergence situation is dangerous because the initial velocity point (V_1, V_2) is in the region of dangerous velocities.

Active braking is selected for the divergence manoeuvre of the first and second UAVs. The choice of the evasion speed is carried out interactively, and the predicted value of the distance of the closest approach is indicated. First, the evasion speed $V_{1y} = 14.5$ knots was introduced for the first UAV. Then the second UAV is selected, and the input of the evasion speed is completed when the distance of the closest approach reaches the maximum permissible distance (0.99 miles), as shown in Figure 9. At the same time, the speed of the second UAV is $V_{2y} = 6.5$ knots. The point corresponding to the selected evasion speeds $V_{1y} = 14.5$ knots and $V_{2y} = 6.5$ knots is located near the boundary of the dangerous speeds area, and the evasion speeds are shown by dots on the UAV braking curves displayed on the right side of the monitor screen.

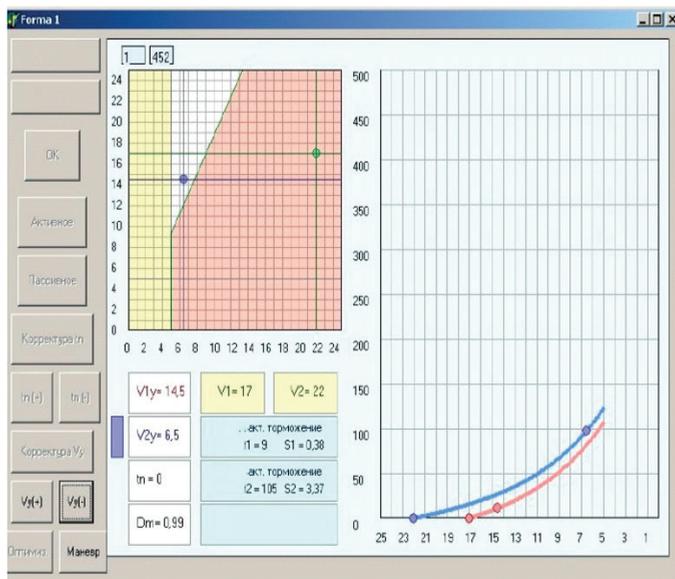


Figure 9. Selection of optimal UAV divergence speeds

Thus, with both types of optimal UAV divergence manoeuvres, the optimality criterion is the loss of the UAV's running time to perform the divergence manoeuvre, which must be minimised. This is expressed as follows. From the expression (1) for $\min D$ we get:

$$K_{ot} = \alpha \mp \arcsin\left(\frac{d_d}{D}\right) \dots\dots\dots (21)$$

therefore, equality is fair:

$$tgK_{ot} = tg\left[\alpha \mp \arcsin\left(\frac{d_d}{D}\right)\right] \dots\dots\dots (22)$$

Denoting it:

$$\gamma^{(1,2)} = \alpha \mp \arcsin\left(\frac{d_d}{D}\right) \dots\dots\dots (23)$$

therefore:

$$tgK_{ot} = tg\gamma^{(1,2)} \dots\dots\dots (24)$$

Taking into account:

$$tgK_{ot} = \frac{V_1 \sin K_1 - V_2 \sin K_2}{V_1 \cos K_1 - V_2 \cos K_2} \dots\dots\dots (25)$$

we will get:

$$\frac{V_1 \sin K_1 - V_2 \sin K_2}{V_1 \cos K_1 - V_2 \cos K_2} = \frac{\sin \gamma^{(1,2)}}{\cos \gamma^{(1,2)}} \quad (26)$$

therefore, we write down the relationship between the course of one UAV K_1 and the speed of another UAV V_2/V_1 , which satisfy the condition $\min D = d_d$. Expression (74) takes the following form:

$$\sin K_1 \cos \gamma^{(1,2)} - \cos K_1 \sin \gamma^{(1,2)} = \frac{V_2}{V_1} (\sin K_2 \cos \gamma^{(1,2)} - \cos K_2 \sin \gamma^{(1,2)}) \quad (27)$$

or:

$$\sin(K_1 - \gamma^{(1,2)}) = \frac{\sin(K_2 - \gamma^{(1,2)})}{V_1} V_2 \quad (28)$$

If denote:

$$\mu^{(1,2)} = \frac{\sin(K_2 - \gamma^{(1,2)})}{V_1} \quad (29)$$

that:

$$\frac{\sin(K_1 - \gamma^{(1,2)})}{\mu^{(1,2)}} \quad (30)$$

So, there are two limits on which the equality $\min D = d_d$ is achieved:

$$V_2^{(1)} = \frac{\sin(K_1 - \gamma^{(1)})}{\mu^{(1)}} = \frac{V_1}{\sin[K_2 - (\alpha + \arcsin \frac{d_d D}{D})]} \sin \left[K_1 - \left(\alpha - \arcsin \frac{d_d D}{D} \right) \right] \quad (31)$$

$$V_2^{(2)} = \frac{\sin(K_1 - \gamma^{(2)})}{\mu^{(2)}} = \frac{V_1}{\sin[K_2 - (\alpha - \arcsin \frac{d_d D}{D})]} \sin \left[K_1 - \left(\alpha + \arcsin \frac{d_d D}{D} \right) \right] \quad (32)$$

Since we are considering a change in speed by braking, the speed values $V_2^{(1,2)}$ must satisfy the condition $V_{2n} > V_2^{(1,2)} \geq 0$

where: V_{2n} is the initial speed of the UAV, which changes its speed with a difference. Let's consider which values of the course K_1 correspond to the limit values 0 and V_{2n} of the speed $V_2^{(1,2)}$. First of all, we note that the boundaries cannot be defined for a situation where:

$$K_2 = \alpha + \arcsin \frac{d_d}{D} \quad (33)$$

Obviously, from the boundary equations (33):

$$K_1 \left(V_2^{(1,2)}(0) = \arcsin \frac{d_d}{D} = \gamma^{(1,2)} \right) \quad (34)$$

To determine the second value:

$$K^1(V_2^{(1,2)} = V_{2n})$$

consider the equation:

$$V_2^{(1,2)} = \frac{\sin(K_1 - \gamma^{(1,2)})}{\mu^{(1,2)}} \quad (35)$$

and substitute $V_2^{(1,2)} = V_{2n}$:

$$V_{2n} = \frac{\sin(K_1 - \gamma^{(1,2)})}{\mu^{(1,2)}} \quad (36)$$

where from:

$$\sin(K_1 - \gamma^{(1,2)}) = \frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(1,2)}) \quad (37)$$

therefore:

$$K_1 = \gamma^{(1,2)} + \arcsin \left[\frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(1,2)}) \right] \quad (38)$$

or:

$$K_1(V_2^{(1,2)} = V_{2n}) = \gamma^{(1,2)} + \arcsin \left[\frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(1,2)}) \right] \quad (39)$$

In the case of $V_1 > V_{2n}$, there are values:

$$K_1(V_2^{(1,2)} = V_{2n}) \quad (40)$$

for both limits, and if $V_1 < V_{2n}$, then it is necessary to take into account the ratio between the magnitude $\gamma^{(1,2)}$ and the extreme relative rates K_{otmax} and K_{otmin} .

Let's consider the case when $V_1 > V_{2n}$ and find the limiting values of the course of the first UAV:

$$K_1(V_2^{(1)} = 0) = \alpha - \arcsin \frac{d_d}{D} = \gamma^{(1)} \quad (41)$$

$$K_1(V_2^{(2)} = 0) = \alpha + \arcsin \frac{d_d}{D} = \gamma^{(2)} \quad (42)$$

$$K_1(V_2^{(1)} = V_{2n} = \gamma^{(1)} + \arcsin \left[\frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(1)}) \right]) \quad (43)$$

$$K_1(V_2^{(2)} = V_{2n} = \gamma^{(2)} + \arcsin \left[\frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(2)}) \right]) \quad (44)$$

Let's introduce the designation:

$$K_{1min}^{(1)} = K_1(V_2^{(1)} = 0), K_{1min}^{(2)} = K_1(V_2^{(2)} = 0) \quad (45)$$

$$K_{1min}^{(1)} = K_1(V_2^{(1)} = 0), K_{1min}^{(2)} = K_1(V_2^{(2)} = 0) \quad (46)$$

Taking into account the accepted designations:

$$K_{1min}^{(1)} = \alpha - \arcsin \frac{d_d}{D} = \gamma^{(1)} \quad (47)$$

$$K_{1min}^{(2)} = \alpha + \arcsin \frac{d_d}{D} = \gamma^{(2)} \tag{48}$$

$$K_{1max}^{(1)} = \gamma^{(1)} + \arcsin \left[\frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(1)}) \right] \tag{49}$$

$$K_{1max}^{(2)} = \gamma^{(2)} + \arcsin \left[\frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(2)}) \right] \tag{50}$$

We draw the reader's attention to the fact that the change in the speed of the second UAV V_2 in the section $V_2 \in (0, V_{2n})$ for the first limit occurs on the course section $K_1 \in (K_{1min}^{(1)}, K_{1max}^{(1)})$, that is, on the interval:

$$\Delta K_1^{(1)} = K_{1max}^{(1)} - K_{1min}^{(1)} \tag{51}$$

or taking into account the expressions received:

$$\Delta K_1^{(1)} = \arcsin \left[\frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(1)}) \right] \tag{52}$$

Similarly, for the second limit: $K_1 \in (K_{1min}^{(2)}, K_{1max}^{(2)})$:

$$\Delta K_1^{(2)} = \arcsin \left[\frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(2)}) \right] \tag{53}$$

In turn, the intervals $\Delta K_1^{(1)}$ and $\Delta K_1^{(2)}$ are smaller $\pi/2$, so at these intervals, the value of the velocity V_2 for both limits increase monotonically. Taking into account the results obtained, the area of Ω_{KV} dangerous parameters of the course of one UAV and the speed of the second UAV, bounded by the first G_{1KV} and the second G_{2KV} limits for the case $G_{1KV}V_1 > V_{2n}$ and looks as shown in Figure 10. Thus, if the point with the initial parameters of the UAV motion $M_n(K_{n1}, V_{2n})$ is located between the first G_{1KV} and the second G_{2KV} limits, that is $(K_{n1}, V_{2n}) \in \Omega_{KV}$, then the inequality $\min D(K_{n1}, V_{2n}) < d_d$, and the approach of the UAV is dangerous. In this case, you must select options evasion UAV K_{1y} and V_{2y} , so that the corresponding point $M_y(K_{1y}, V_{2y})$ was closest to the point of $M_n(K_{n1}, V_{2n})$ of scope and the distance between the points M_n and M_y was minimal, as shown in Figure 10.

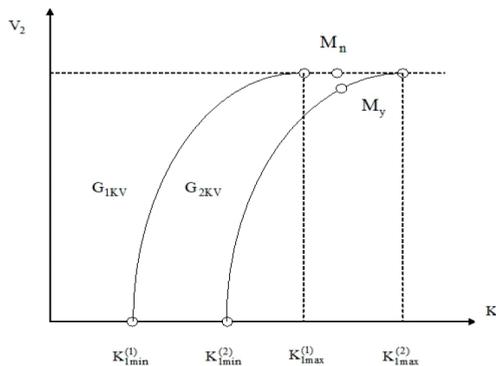


Figure 10. The area of dangerous parameters of the UAV

To calculate the evasion rate of the second UAV, which provides the distance of the closest approach equal to the maximum permissible approach distance for the selected evasion course of the first UAV, analytical expressions should be used for the duration of transients, the distance S travelled during this time and the dependence of the UAV speed on time in cases of active and passive braking. To do this, we will use the results of the work that enables us to obtain dependencies for active braking:

$$\tau_a = \frac{m}{\mu V_{yp}} \left[\arctg \frac{V_0}{V_{yp}} - \frac{V_y}{V_{yp}} \right] \quad (54)$$

$$S_a = \frac{m}{2\mu} \ln \left| \frac{V_0^2 + V_{yp}^2}{V_y^2 + V_{yp}^2} \right| \quad (55)$$

$$V_a = V_{yp} t g \left(\arctg \frac{V_0}{V_{yp}} - \frac{\mu V_{yp}}{m} t \right) \quad (56)$$

Similarly, for passive braking mode:

$$\tau_p = \frac{m}{\mu} \left(\frac{1}{V_y} - \frac{1}{V_0} \right) \quad (57)$$

$$\tau_p = \frac{m}{\mu} \left(\frac{1}{V_y} - \frac{1}{V_0} \right) \quad (58)$$

$$S_p = \frac{m}{2\mu} \ln \left| \frac{V_0^2}{V_y^2} \right| \quad (59)$$

Previously, a procedure was obtained for calculating the boundaries of the region Ω_{KV} in the case of the advantage of the speed of the first UAV ($V_1 > V_{2n}$). If the speed of the second UAV exceeds the speed of the first one, which manoeuvres a change of course, that is, ($V_1 < V_{2n}$), we get the equality:

$$K_1 = \gamma^{(1,2) + \arcsin \left[\frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(1,2)}) \right]} \quad (60)$$

the analysis of which shows that the boundary of the region Ω_{KV} exists for the velocities of the second UAV when the expression under the arcsin function:

$$\left| \frac{V_{2n}}{V_1} \sin(K_2 - \gamma^{(1,2)}) \right| \leq 1 \quad (61)$$

Therefore, the maximum value of the course of the first UAV K_{1max} of the boundary of the region is Ω_{KV} achieved at a speed of:

$$V_{2m} = \frac{V_1}{\sin(K_2 - \gamma^{(1,2)})} \quad (62)$$

and is equal to:

$$K_{1max} = \gamma^{(1,2)} \pm \arcsin(1) = \gamma^{(1,2)} \pm \frac{\pi}{2} \quad (63)$$

So:

$$K_{1max}^{(1,2)} = \gamma^{(1)} - \frac{\pi}{2}, K_{1max}^{(2)} = \gamma^{(2)} + \frac{\pi}{2} \quad (64)$$

Thus, for $V_1 < V_{2n}$, the boundaries of the area Ω_{KV} exist for:

$$K_1 \in \left[\gamma^{(1,2)} - \frac{\pi}{2}, \gamma^{(1,2)} + \frac{\pi}{2} \right] \text{ and } V_2 \in \left[0, \frac{V_1}{\sin(K_2 - \gamma^{(1,2)})} \right] \quad (65)$$

For the development of the area Ω_{KV} and its graphical display, a computer programme was developed that takes into account the ratio of the initial speeds of the UAV V_1 and V_{2n} . In the case of $V_1 > V_{2n}$, there are values:

$$K_1 \left(V_2^{(1,2)} = V_{2n} \right) \quad (66)$$

for both limits, and the area of Ω_{KV} dangerous parameters of the course of one UAV.

The scientific result of solving the second auxiliary task was the methods of developing dangerous areas of UAV movement at the distance of the closest approach.

Conclusion. The article explores the principle of developing a GIS module for identifying dangerous parameters in the routes of multiple UAVs operating within a smart city environment. It demonstrates how this GIS-based approach can be applied to select safe divergence maneuvers for multiple UAVs simultaneously. The concept of situational disturbance in the context of UAV swarms in urban airspace is introduced, along with methods for assessing danger levels. For scenarios involving multiple UAVs, a situational disturbance matrix is employed within the GIS framework. An analysis of various control types for UAV divergence processes in smart city environments is conducted. The study highlights that while traditional collision avoidance relies on locally independent control with binary coordination systems (similar to maritime regulations), modern UAV motion control systems in smart cities utilize the principle of complete external control facilitated by GIS. For external control of UAV swarm divergence in urban settings, the article presents a GIS-based method for assessing convergence situations among multiple UAVs and selecting appropriate divergence maneuvers. This method incorporates real-time spatial data analysis, considering factors such as UAV turning inertia and the presence of urban navigational hazards within the maneuvering area of the smart city. The GIS integration enables comprehensive monitoring of UAV swarms, providing a robust solution for route management and collision avoidance in the

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CONTENT

<p>A.O.Baisalova, A.V. Dolgoplova, R. Seltmann, E.E. Akbarov, M.A. Mashrapova ROGRAPHIC AND MINERALOGICAL FEATURES OF THE KARAGAILY-AKTAS RARE METAL DEPOSIT (SOUTH KAZAKHSTAN REGION).....</p>	6
<p>V.V. Gerasidi, R.G. Dubrovin, O.I. Kukartseva, I.A. Panfilov, V.V. Tynchenko ANALYSIS OF TECHNICAL OPERATION OF CATERPILLAR ENGINEERING CORPORATION ENGINES IN INDUSTRY.....</p>	23
<p>K. Yelemessov, D. Baskanbayeva, L. Sabirova OPTIMIZATION OF TECHNICAL MEANS AND TECHNOLOGICAL PROCESSES OF GAS COMPRESSION FOR THE MINING INDUSTRY OF THE REPUBLIC OF KAZAKHSTAN.....</p>	36
<p>D. Karaivanov, M.F. Kerimzhanova, M.E. Isametova, N.S. Seiitkazy, G. Turymbetova INVESTIGATION OF KINEMATICS AND POWER OF COMPOSITE PLANETARY GEARS FOR WIND TURBINES.....</p>	47
<p>A.Zh. Kassenov, A.Zh. Taskarina, K.K. Abishev, A.D. Suleimenov, D.D. Alipbayev COMBINED TOOL FOR CUTTING INTERNAL THREADS OF TUBING.....</p>	63
<p>Z.A. Kutpanova, D.O. Kozhakhmetova, G. Baiseitov, A.Dolya, G.A. Uskenbayeva ROUTE CONTROL AND COLLISION AVOIDANCE FOR MULTIPLE UAVS IN A SMART CITY CONTEXT USING GEOGRAPHIC INFORMATION SYSTEM.....</p>	71
<p>I.Yu. Matasova, Yu.S. Kuznetsova, T.A. Panfilova, V.S. Tynchenko, S.V. Tynchenko. FEATURES OF THE BEHAVIOR OF ROCKS IN THE UNDERGROUND FIELD DEVELOPMENT.....</p>	94
<p>M. Nurpeisova, B. Mingzhasarov, K. Temirkhanov, Y. Kakimzhanov, Zh. Nukarbekova GEODETIC MONITORING OF DEFORMATION PROCESSES AT KAPCHAGAY HYDROPOWER PLANT.....</p>	107

L. Nurshakhanova, S. Zakenov, A. Zakenova TECHNOLOGIES OF WATER-GAS IMPACT ON THE RESERVOIR USING SIMULTANEOUSLY PRODUCED PETROLEUM GAS.....	118
T.K. Salikhov, A.I. Abekeshev, G.O. Abisheva, Zh.B. Issayeva, . M.B. Khussainov STUDY OF THE ECOSYSTEM AND UNIQUE NATURAL OBJECTS OF THE CHINGIRLAU DISTRICT OF THE WEST KAZAKHSTAN REGION USING GIS TECHNOLOGIES.....	128
V.V. Sirota, S.V. Zaitsev, M.V. Limarenko, D.S. Prokhorenkov, A.S. Churikov THERMOMECHANICAL PROCESSING OF MINERAL RAW MATERIALS TO PRODUCE $La_{1-x}Sr_xMn_zO_3$ POWDER WITH PEROVSKITE STRUCTURE.....	155
A.V. Taranov, A.D. Mekhtiyev, F.N. Bulatbayev, Y.G. Neshina, V.S. Balandin PNEUMATIC LOAD HOISTS FOR MINERAL TRANSPORTATION FROM MINES.....	167
Y.A. Tynchenko, B.V. Malozyomov, V.V. Kukartsev, M.A. Modina, G.L. Kozenkova APPLYING ELEMENTS OF A TOTAL EQUIPMENT CARE STRATEGY TO ANALYZE THE OPERATION OF MINING MACHINERY.....	178
O.G. Khayitov, J.B. Toshov, K.T. Sherov, B.N. Absadykov, M.R. Sikhimbayev OIL AND GAS POSSIBILITY OF THE CENTRAL GRABEN OF THE BUKHARA-KHIVA PALEORIFTS AND ITS PERSPECTIVES.....	191
Z.I. Chernenko, M.A. Mizernaya, I. E. Mataibayeva, N.A. Zimanovskaya GOLD ORE DEPOSITS ASSOCIATED WITH CARBONATE FORMATIONS (EAST KAZAKHSTAN).....	201

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